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DEVELOPMENT OF AN EFFICIENT COMPUTER CODE TO SOLVE  
THE TIME-DEPENDENT NAVIER-STOKES EQUATIONS

FINAL REPORT

REPORT NO. SR-26

DECEMBER 1975

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1. SUMMARY

A research effort was conducted with the goal of reducing computer time of a Navier Stokes Computer Code for prediction of viscous flow fields about lifting bodies. A two-dimensional, time-dependent, laminar, transonic computer code (STOKES) was modified to incorporate a non-uniform timestep procedure. The non-uniform timestep requires updating of a zone only as often as required by its own stability criteria or that of its immediate neighbors. In the uniform timestep scheme each zone is updated as often as required by the least stable zone of the finite difference mesh. Because of less frequent update of program variables it was expected that the non-uniform timestep would result in a reduction of execution time by a factor of five to ten. Available funding was exhausted prior to successful demonstration of the benefits to be derived from the non-uniform timestep method.

## 2.0 INTRODUCTION

Two elements lie at the core of any satisfactory numerical procedure for computing laminar and/or turbulent transonic airfoil flows, namely, (a) a general method for integrating the equations of continuum mechanics, to which equations governing the evolution of turbulence variables can be added, and (b) a mathematical model that describes the fluctuating component of an arbitrary turbulent field. To the extent possible, both the numerical method and the turbulence model should offer proven reliability; the choices proposed are believed to best meet such a criterion, namely:

1. A flow field computer code, which integrates the Navier-Stokes equations in two spatial dimensions, and
2. The Saffman Model of turbulent flow.

For each of the many laminar flows (References 1-7) to which it has been applied, the presently operational flow field computer code has been shown to yield solutions accurate enough for any engineering purpose. Furthermore, in the two-dimensional test calculations completed to date, the less proven Saffman turbulence (Ref. 8-9) model, in combination with the flow field computer code, has provided an accurate solution to the turbulent flat plate problem, (Ref. 10) a turbulent boundary layer-shock wave interaction (Ref. 10) and the separation phenomena that occur on a compression corner in supersonic flow. (Ref. 10)

A version of the Afton plane symmetric code (A2P) was developed for the NASA Ames Research Center to solve a laminar transonic airflow problem. The flow field was computed about the lifting NACA 64A410 airfoil at Mach .72, Reynolds number (based on chord length) of  $1.75 \times 10^6$  and at an angle of attack of  $4^\circ$  (ref 11). Starting from the impulsive conditions the calculation was carried out to the time it takes a freestream particle to travel one chord length. The calculation required slightly less than five hours of CDC 7600 computer time. In the course of the calculation the flow separated on the leeward trailing edge, a leeward compression wave formed, and approximately 50% of the measured lift was computed.

However, it was estimated that an additional ten hours on the CDC 7600 computer would be required to complete the problem.

Computing times of the order of fifteen hours on the CDC 7600 to complete one transonic airfoil calculation make transonic flow computations about lifting bodies impractical for design purposes. A speed-up factor of the order of 10 would be necessary for numerical computation of transonic flow to be in the practical range. Coupled with a better estimate of the initial conditions than impulsive initial conditions, and with a speed-up factor of ten, transonic lifting airfoil problems can be solved in less than an hour on the CDC 7600 computer.

The main objective of the work conducted in this program was to develop a computer code which would be approximately an order of magnitude faster than the present computer program. This was to be accomplished by use of a "Non-Uniform Timestep" in the computation procedure to improve computation efficiency.



### 3.0     SYMBOLS

E	Specific internal energy
j	Index specifying streamline-like direction
k	Index specifying potential-like direction
M	Mass
n	Time index for finite difference equation
p	Pressure
$\Delta t$	Timestep increment

#### 4.0 NON-UNIFORM TIMESTEP FOR IMPROVMENT IN COMPUTATIONAL EFFICIENCY OF NAVIER STOKES SOLUTIONS

The STOKES computer code was developed for the NASA Ames Research Center from an AFTON plane symmetric code (A2P). It is a two-dimensional time-dependent code used to predict laminar transonic flows about lifting bodies. In common with other AFTON codes, a uniform timestep was incorporated in the original version of the program. The uniform timestep procedure requires that all of the variables of motion in every zone of the mesh be updated or calculated as often as required by stability considerations for the least stable zone of the mesh. The timestep is dependent upon the time required for the speed of sound to cross a zone. It was anticipated that significant computer time savings would be achieved if the non-uniform timestep procedure were incorporated in the code. This allowed the variables of motion to be calculated for a give zone of the finite difference mesh only as often as required by considerations of stability of that zone and its immediate neighbors. It was estimated that a speed-up factor between five and ten would result from the incorporation of non-uniform timestep procedure.

##### 4.1 Treatment of Timestep Intermesh Boundaries

One of the most crucial aspects of the non-uniform timestep procedure relates to the way boundaries between adjacent zones having different timesteps are treated. In order to maintain AFTON conservation properties across the interface it had been previously determined by Trulio (Ref. 12) that correct time-centering of the pressures used to update the momenta was essential. As initially conceived, the interface treatment would assume that the adjoining large  $\Delta t$  zone

would be treated in nearly all respects as the small timestep level zone (a micro zone) and updated with a small timestep. In this macro boundary zone the explicitly calculated pressure available from the first microstep update (there would be two such calculations in order to bring the zone correctly forward in time equivalent to a single macrostep calculation) is the correct pressure to use in updating the adjacent momentum zones. This pressure would be used for both microstep momentum updates and the single macrostep momentum update. This procedure would have correctly treated the momentum conservation constraints in the transition from a micro zone to a macro zone.

The method described above does not correctly conserve the total energy in the same transition region. For a macro level zone,  $j$ , in the Lagrangian case,

$$M_j (E_j^{n+1} - E_j^{n-1}) = -P_j^n \Delta V_j^n \quad (1)$$

must be true to conserve total energy; where  $M_j$  is the zone mass,  $E_j$  the specific internal energy,  $P_j$  the zone pressure and  $\Delta V_j$  the change in volume. Superscripts refer to centering in time;  $n$  is a microstep increment. Macrozone energy is updated in time from  $n - 1$  to  $n + 1$ , an interval of  $2n$  or two microsteps. Had we treated this boundary macro zone as a micro zone, as described in Paragraph 2, two such updates would be required.

$$M_j (E_j^n - E_j^{n-1}) = P_j^n \Delta V_j^n \quad (2)$$

$$M_j (E_j^{n+1} - E_j^n) = -P_j^n \Delta V_j^n \quad (3)$$

The pressure would no longer be correctly centered in relation to the internal energy. The resulting change in internal energy from  $E_j^{n-1}$  to  $E_j^{n+1}$  would not be the same as that calculated by Equation (1).

The interface was revised to incorporate a different method which preserves the general conservation properties. Macro zones in the interfaces were to be treated as macro zones in general respects; an intermediate pressure centered at  $n$  is calculated in the macro zone and used to update the adjoining momentum zones. This pressure at  $n$  is used with Equation (1) to explicitly calculate the internal energy at  $n + 1$  and so preserve total energy conservation. Logic and coding were developed and inserted in the updated program.

#### 4.2 Development of Non-Uniform Timestep Logic

The logic for controlling the zone properties updating sequence and the assignment of timestep levels for thermodynamic and momentum zones was included in two new subroutines inserted in the code. These subroutines, ZONEL and TMESTP are shown in the program flow diagram presented in Figure 1. ZONEL is the routine which assigns timestep levels to the thermodynamic and momentum zones. It is called from FLOW at the beginning of a new macrocycle. At that point all zones have been updated to the same time and new levels must be assigned for the new updating sequence.

The TMESTP subroutine calculates and stores the actual timestep, based upon stability criteria, for each zone. It replaces coding formerly in MVS. This routine is also called from FLOW, but after MVS and after the final macrolevel update. The timesteps are then available for ZONEL to perform its required testing and assignment at the beginning of the next macrocycle. A listing of subroutine ZONEL is included as an Appendix to this report.

The program, as written, is currently restricted to only two timestep levels, although additional time reduction should result from multiple time levels. Thus, zones which are at least twice as large as the smallest zone require updating only every other cycle. Zones which are four times larger than the smallest zone would normally require updating only every fourth cycle, etc. A temporary restriction in timestep levels was inserted to simplify checkout and debugging. Coding is in place which will eventually allow up to five timestep levels.

#### 4.3 Development of Conservative Logic

In order to maintain the conservative aspects of the AFTON numerical method an approach was taken which utilized momentum and mass accumulators to determine mesh boundary fluxes. This approach, because of its inherent simplicity, offered substantial savings in execution time in addition to the savings which were expected to result from the non-uniform timestep.

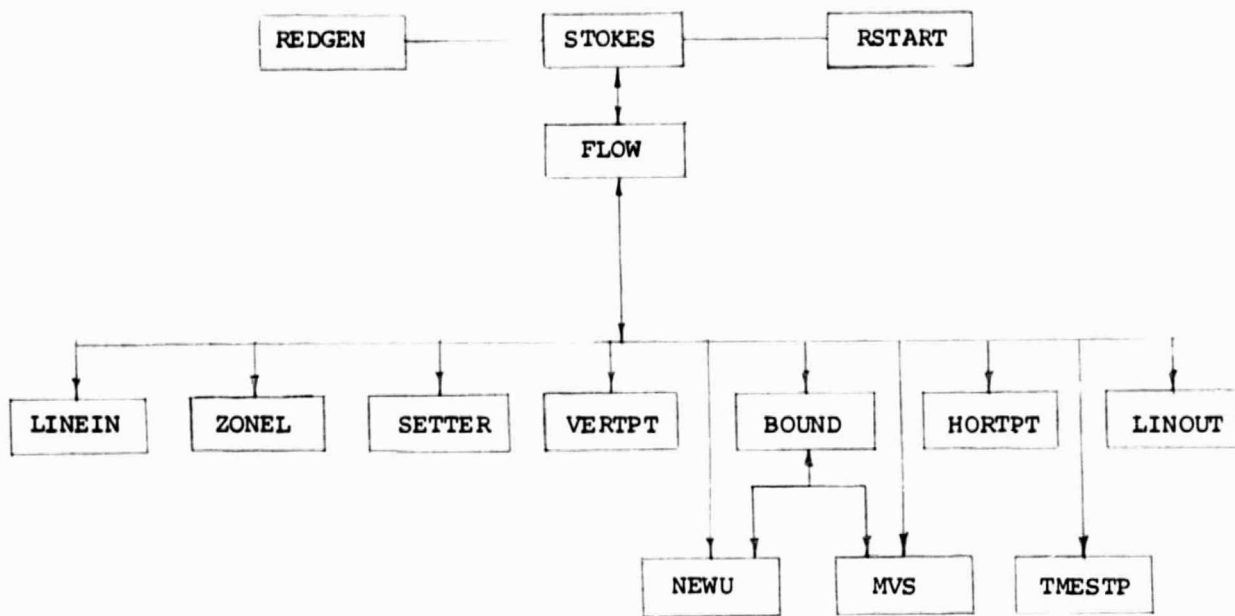


Figure 1: General Flow Diagram for Computer Program With Non-Uniform Timestep Scheme.

Complete conservation also required calculation of the work done on the surfaces of any body placed in the flow field. The AFTON differencing scheme results in a half momentum zone around each of the mesh boundaries and around any body in the mesh. Logic had to be implemented which accounted for the fact that the work calculation for the body surfaces included the momentum transported across these half-zone boundaries.

The various difficulties which were encountered while incorporating the conservation logic eventually led to exhaustion of the available funding for this program. It is estimated that about one-half of the conservation logic had been inserted in the code. Substantial additional programming and debugging effort is required to complete this task and insure that all of the conservation checks of the AFTON method are preserved.

#### 4.4 Checkout and Debugging of Non-Uniform Timestep Coding

A test problem was run to check out the code for laminar flow over a right circular cylinder. Three cycles were run with timestep restricted to two timestep levels. The program appeared to execute normally. However, validation of the changes requires completion of the conservation logic and coding.

## 5.0 CONCLUDING REMARKS

An effort was undertaken to incorporate a non-uniform timestep in a two-dimensional, time-dependent, Navier-Stokes solution of the laminar, transonic flow over a lifting body. Available funding was exhausted during the implementation of the conservation logic. Without the conservation properties the accuracy of the numerical calculation is uncertain. Successful demonstration of the non-uniform timestep as a method for improving the efficiency of Navier-Stokes solutions requires additional programming and debugging effort.

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# APPENDIX

## Subroutine ZONEL

ZONEL

SUBROUTINE ZONEL

ZONEL.2

ZONEL.3

```

C
COM* FCOPY
VC
PARAMETER MJ=68,MQ=83+295*MJ,IQ=28*MJ
COMMON SAV(6),TIME,SMOMXI,SMXTPT,SMOMX,SMOMYI,SMYTP,SMOMY,SENERI,
1 SIETPT,SKETPT,MURK,SUMIE,SUMKE,SUMTE,FIMPX,FIMPY,SMASSI,SMSTPT,
2 SMASS,PROHNO,VZ,GAMMA,EMU,RHJIN,UXIN,UYIN,EIN,DTNM,CUTOFF,NJPA,
3 NECIT,NJMIN,JMAX,KMIN,KMAX,KBOT,KTOP,MAXN,JR,KT,TMAX,JB,JBM,
4 JBM2,KSV(12),NSIG,DTNMN,KMT
COMMON A(68), B(68,4), RX(68,5), RY(68,5), UNMX(68,5),
1 UNPY(68,5), UNPX(68,5), UNPY(68,5), FMASNM(68,5), FMASN(68,5),
2 ENM(68,5), EN(68,5), PNM(68,5), PN(68,5), PSNXX(68,5),
3 PSNMY(68,5), PSNMY(68,5), PSVXX(68,5), PSVXY(68,5), PSVYY(68,5),
4 RAA3Z(68,5), RAA1Z(68,5), RAAE3Z(68,5), RAAE1Z(68,5), RAA3Z(68,5),
5 RAA1Z(68,5), E3Z(68,5), E1Z(68,5), RHJ(68,5), VOL(68,5),
6 ETA(68,5), AZX(68,5), A1X(68,5), A2X(68,5), AX(68,5),
7 AZY(68,5), A1Y(68,5), A2Y(68,5), ASY(68,5), ARX(68,5),
8 ARY(68,5), ALX(68,5), ALY(68,5), DJDX(68), DJDY(68),
9 DVDX(68), DVDY(68), SIGX(68,5), SIGY(68,5), TAUXY(68,5)
COMMON
1 FRNMX(68,5), FRNMY(68,5), FLNMX(68,5), FLNMY(68,5), FMSNZ(68,5),
2 FMNMX(68,5), FMNMY(68,5), NTPT(68,5), FNMX(68,5), FMNY(68,5)
COMMON LX1, LX2, LX3, LX4,
1 LX5, LINCT, NMAS, ICUN,
2 KC, GAMFAC, EMU2, DTNMP5,
3 DTNM2, CUT1, CUT2, CUT,
4 NDMP
COMMON
1 LEVMX, LV1, LV2, LV3, LV4, LV5, MAXLEV,
2 MICRO,
*
1 ACE(68,5), ACM(68,5), ACXM(68,5), ACYM(68,5), DTLEV(10),
2 JIF(68,5), JML(68,5), JZL(68,5), ZOT(68,5)
C
COMMON NMDD
COMMON /SBLK/ S(36,68),SS(36,68),
1 NTS,NTSS,MINS,MOUTS,NT,MT,MJ,MQ,IQ,MHINS,MMDOTS,KBLK,KKBLK
LARGE A(20000), W(20000)
C=
C = LX3
2 L5 = LX2
4 IF(KC,GT,1) GO TO 20
10 JMP = JMIN + 1
11 MAXLEV = 2
13 DO 5 M=1,10
14 IF(M,GT,MAXLEV) GO TO 4
17 DTLEV(M) = DTNM*(2.**(M-1))
25 GO TO 5
25 4 DTLEV(M) = 1.
27 5 CONTINUE
31 LCT = 0
32 L = LX1
33 L2 = LX2
35 GO TO 20
36 10 CONTINUE
36 L = LX2

```

ZONEL.5

ZONEL.6

ZONEL.7

ZONEL.8

ZONEL.9

ZONEL.10

ZONEL.11

ZONEL.12

ZONEL.13

ZONEL.14

ZONEL.15

ZONEL.16

ZONEL.17

ZONEL.18

ZONEL.19

ZONEL.20

ZONEL.21

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ZONEL	37	L5 = LX1	ZONEL,22
	41	GO TO 20	ZONEL,23
	42	15 CONTINUE	ZONEL,24
	42	L = LX3	ZONEL,25
	43	L5 = LX2	ZONEL,26
	45	20 CONTINUE	ZONEL,27
	45	LCT = LCT + 1	ZONEL,28
	47	DO 50 J = JMIN,JR	ZONEL,29
	51	IF(J,EG,JBM) GO TO 50	ZONEL,30
	52	IF(ZDT(J,L),GT,DTLEV(2)) GO TO 25	ZONEL,31
	60	JZL(J,L) = 1	ZONEL,32
	61	GO TO 50	ZONEL,33
	C	TEMPORARY COMMENT OUT OF TRANSFER TO LIMIT TO TWO LEVELS	ZONEL,34
	C25	IF(ZDT(J,L),GT,DTLEV(3)) GO TO 30	ZONEL,35
	62	25 CONTINUE	ZONEL,36
	62	JZL(J,L) = 2	ZONEL,37
	66	GO TO 50	ZONEL,38
	67	30 IF(ZDT(J,L),GT,DTLEV(4)) GO TO 35	ZONEL,39
	75	JZL(J,L) = 3	ZONEL,40
	76	GO TO 50	ZONEL,41
	77	35 IF(ZDT(J,L),GT,DTLEV(5)) GO TO 40	ZONEL,42
	105	JZL(J,L) = 4	ZONEL,43
	106	GO TO 50	ZONEL,44
	107	40 CONTINUE	ZONEL,45
	107	JZL(J,L) = 5	ZONEL,46
	113	50 CONTINUE	ZONEL,47
	116	IF(LCT,EG,1) GO TO 10	ZONEL,48
	120	JML(JMIN,L) = MIN0(JZL(JMIN,L),JZL(JMIN,L5))	ZONEL,49
	132	DO 60 J = JMP,JR	ZONEL,50
	133	IF(J,EG,JR,OK, J,EG,JBM) GO TO 60	ZONEL,51
	155	JM = J - 1	ZONEL,52
	156	JML(J,L) = MIN0(JZL(JM,L),JZL(J,L),JZL(JM,L5),JZL(J,L5))	ZONEL,53
	167	60 CONTINUE	ZONEL,54
	172	JML(JMAX,L) = MIN0(JZL(JR,L),JZL(JR,L5))	ZONEL,55
	204	IF(JB,GT,JMAX) GO TO 65	ZONEL,56
	207	IF(KBDT=LCT) 61,62,65	ZONEL,57
	211	61 IF(LCT=KTOP) 63,64,65	ZONEL,58
	C		ZONEL,59
	C	SPECIAL TESTING FOR KBDT	ZONEL,60
	C		ZONEL,61
	214	62 CONTINUE	ZONEL,62
	214	J1= JR	ZONEL,63
	215	J2 = JR - 2	ZONEL,64
	217	J3 = JB	ZONEL,65
	220	621 CONTINUE	ZONEL,66
	220	JML(J1,L) = MIN0(JZL(J3,L),JZL(J2,L5),JZL(J3,L5))	ZONEL,67
	240	IF(J1,NE,JR) GO TO 60	ZONEL,68
	242	J1= JRM	ZONEL,69
	243	J2 = JRM + 1	ZONEL,70
	244	J3 = JRM - 1	ZONEL,71
	245	GO TO 621	ZONEL,72
	C		ZONEL,73
	C	SPECIAL TESTING FOR KLINES AROUND THE BODY	ZONEL,74
	C		ZONEL,75
	245	63 CONTINUE	ZONEL,76
	245	J1= JB	ZONEL,77

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ZONFL				
246		J2= JB		ZONEL,75
250	631	CONTINUE		ZONEL,79
250		JML(J1,L) = MINO(JZL(J2,L),JZL(J2,L5))		ZONEL,80
263		IF(J1,NE,JB) GO TO 66		ZONEL,81
265		J1= JBM		ZONEL,82
266		J2 = JBM - 1		ZONEL,83
267		GO TO 631		ZONEL,84
	C			ZONEL,85
	C			ZONEL,86
	C	SPECIAL TESTING FOR KTDP		ZONEL,87
270	64	CONTINUE		ZONEL,88
270		J1 = JB		ZONEL,89
271		J2 = JB - 2		ZONEL,90
273		J3 = JB		ZONEL,91
274	641	CONTINUE		ZONEL,92
274		JML(J1,L) = MINO(JZL(J3,L),JZL(J3,L5),JZL(J2,L))		ZONEL,93
313		IF(J1,NE,JB) GO TO 66		ZONEL,94
315		J1= JBM		ZONEL,95
316		J2= JBM + 1		ZONEL,96
317		J3= JBM - 1		ZONEL,97
320		GO TO 641		ZONEL,98
320	65	CONTINUE		ZONEL,99
320		J1= JB		ZONEL,100
321		J2= JB - 2		ZONEL,101
323		J3 = JB		ZONEL,102
324	651	CONTINUE		ZONEL,103
324		JML(J1,L) = MINO(JZL(J2,L),JZL(J3,L),JZL(J2,L5),JZL(J3,L5))		ZONEL,104
351		IF(J1,NE,JB) GO TO 66		ZONEL,105
353		J1= JBM		ZONEL,106
354		J2 = JBM - 1		ZONEL,107
355		J3 = JBM + 1		ZONEL,108
356		GO TO 651		ZONEL,109
356	66	CONTINUE		ZONEL,110
356		IF(LCT,NE,2) GO TO 68		ZONEL,111
360		L = LX1		ZONEL,112
361		L2 = LX2		ZONEL,113
363		DO 67 J = JMIN,JB		ZONEL,114
374	67	JML(J,L) = JML(J,L2)		ZONEL,115
376		GO TO 69		ZONEL,116
377	68	L = LX2		ZONEL,117
400		L2 = LX3		ZONEL,118
402	69	CONTINUE		ZONEL,119
402		DO 70 J = JMIN,JB		ZONEL,120
404		IF(J,EQ,JBM) GO TO 70		ZONEL,121
420		JP = J + 1		ZONEL,122
421		JTMP = MINO(JML(J,L) , JML(J,L2),JML(JP,L),JML(JP,L2))		ZONEL,123
430		IF(JTMP,EQ,JZL(J,L)) GO TO 70		ZONEL,124
435		JIF(J,L) = 1		ZONEL,125
436		JZL(J,L) = JTMP		ZONEL,126
437	70	CONTINUE		ZONEL,127
442		IF(LCT,EQ,2) GO TO 15		ZONEL,128
444		RETURN		ZONEL,129
444		END		ZONEL,130

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